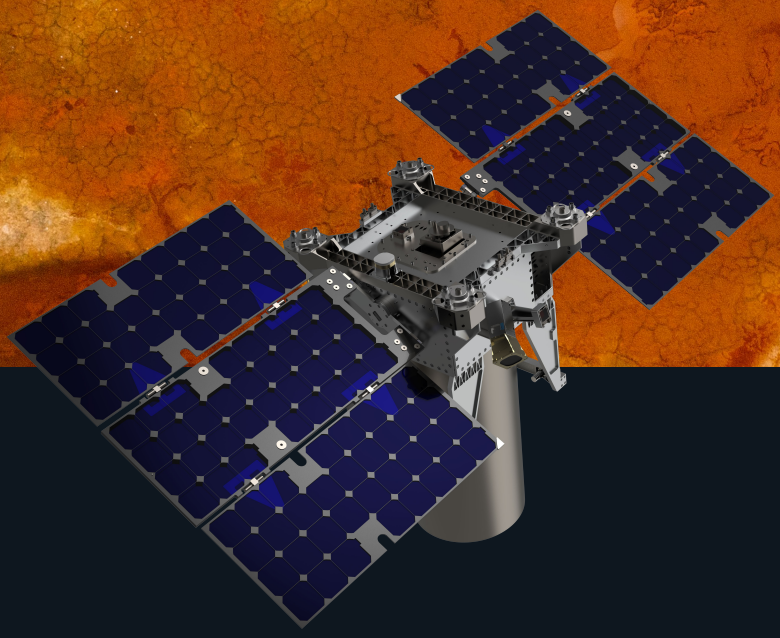
VENUS-100
MICROSATELLITE



CLASS
ESPA-Standard or large 15” launch vehicle interface, 4-point interface available

POINTING ACCURACY
< 5 arcsec (1-sigma), per axis

SOLAR ARRAY POWER (BOL)
Two wing: 444 W
One wing: 222 W

PAYLOAD MASS CAPABILITY
70 kg

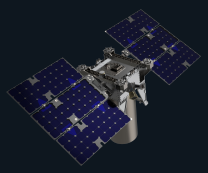
ENERGY STORAGE
13.6 Ah

ORBIT ALTITUDE / ORBIT LIFETIME
LEO (> 7 years), GEO (> 5 years),
Deep Space (> 5 years)

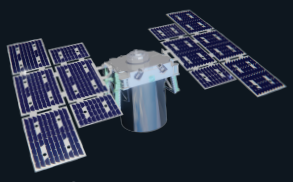
PAYLOAD VOLUME
17.0” X 16.4” X 27.0” (2 array)

Above stowed volumes keep payload within bus payload deck footprint and ESPA/G/Rideshare radial dimensions. Larger stowed volumes can be accommodated depending on launch vehicle configuration. Deployed volumes are CG and MOI dependent.

MICROSAT & MINISAT SUMMARY

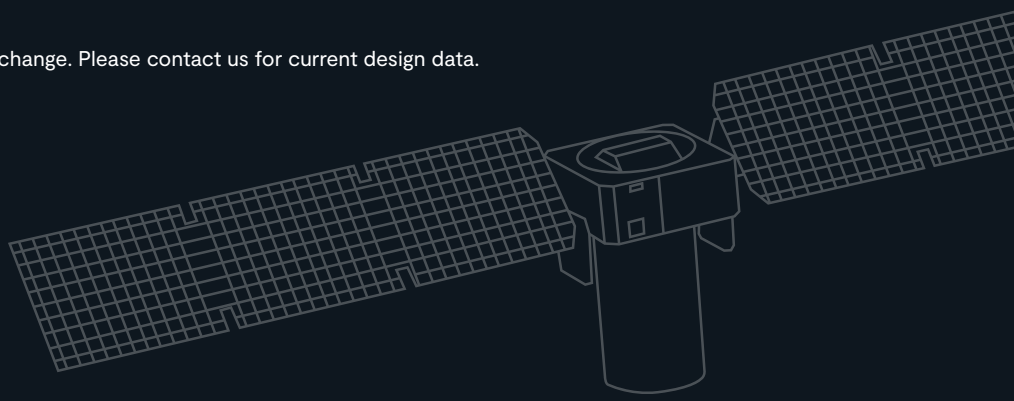






	VENUS-100	SATURN-200	SATURN-400
CLASS	ESPA-Standard or large 15” launch vehicle interface, 4-point interface available	ESPA-Grande or Equivalent 24” launch vehicle interface standard, other options available	Rideshare and NSSL compatible
POINTING ACCURACY	< 5 arcsec (1-sigma), per axis		
SOLAR ARRAY POWER (BOL)	One wing: 222 W Two wing: 444 W	1082 W	1876 W
PAYLOAD MASS CAPABILITY	70 kg	Standard rideshare*: 200 kg Specialty applications and options*: up to 350 kg *CG and MOI dependent	Standard rideshare*: 250 kg (objective) Specialty applications and options*: 600 kg (objective) *CG and MOI dependent
ENERGY STORAGE	13.6 Ah	54.4 Ah	75 Ah Options for increased energy storage available.
ORBIT ALTITUDE / ORBIT LIFETIME	LEO (> 7 years), GEO (> 5 years), Deep Space (> 5 years)		
PAYLOAD VOLUME		30.0” X 30.0” X 40.0” (typical)	40.0” X 40.0” X 45.0” (objective)
	Above stowed volumes keep payload within bus payload deck footprint and ESPA/G/rideshare radial dimensions. Larger stowed volumes can be accommodated depending on launch vehicle configuration. Deployed volumes are CG and MOI dependent.		
PROPULSION	Multiple electric and chemical propulsion systems available		

Note: This data is for information only and subject to change. Please contact us for current design data.



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